

AMENDMENTS TO THE CLAIMS:

This listing of claims will replace all prior versions, and listings, of claims in the application:

☐ 1.-8. (Cancelled).

29 ~~9.~~ (Currently amended) A thermal barrier coating applied to a metallic component of a gas turbine engine comprising a top layer of a dense vertically cracked ceramic consisting of ~~less than 6~~ about 4 weight % Ytria, 0-1 weight % Hafnia, and the balance Zirconia, and a bond coating adhering said ceramic layer to said metallic component, said bond coating comprising an oxidation-resistant alloy of MCrAlY with a thickness of about .003 to .025 inches, where M is iron, cobalt or nickel.

~~2~~ ~~10.~~ (Previously Presented) A thermal barrier coating as recited in claim ~~9~~,¹ wherein said dense vertically cracked ceramic layer is about 5-100 mils thick.

☐ 11. (Cancelled).

~~3~~ ~~12.~~ (Previously Presented) The thermal barrier coating system as recited in claim ~~9~~,¹ wherein said Zirconia is partially stabilized by Ytria.

~~4~~ ~~13.~~ (New) A thermal barrier coating applied to a metallic component of a gas turbine engine comprising a top layer of a dense vertically cracked ceramic consisting of about 4 weight % Ytria, 0-1 weight % Hafnia, and the balance Zirconia, and a bond coating adhering said ceramic layer to said metallic component, said bond coating comprising a diffusion aluminide or a platinum aluminide that forms an oxidation-resistant intermetallic.